



POWER PLANT - ENGINE - INCORPORATE REVISED BRACKET ATTACHING BOLTS AT THE FLANGE FA1 -  
CATEGORY CODE 3 - MOD.ENG-71-0057

Printed in Great Britain

1. Planning Information

A. Effectivity

- (1) Aircraft: Airbus A320
- (2) Engine: V2500-A1 Engines, Serial Number V0014

B. Reason

(1) Condition

The bolts which install the two brackets including the bracket part number 5W8317 to the flange FA1, may have an insufficient length to tighten the two brackets together with the flange FA1.

This could cause a looseness of the bolts, nuts and the brackets because of possible unsatisfactory self locking of the nuts at the bolt ends.

(2) Background

This condition has been noted when the two brackets including the bracket PN 5W8317 are installed to the flange FA1.

A review for the existing configuration has shown that the bolt length is not enough to install the two brackets.

(3) Objective

The changes in configuration recommended in this Service Bulletin are intended to maintain engine reliability.

(4) Substantiation

Substantiation test is not required.

(5) Effect of Bulletin on workshop procedure:

Removal/Installation - Not affected.  
Disassembly/Assembly - Not affected.  
Cleaning - Not affected.  
Inspection/Check - Not affected.  
Repair - Not Affected.  
Testing - Not affected

(6) Supplemental Information

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None.

C. Description

(1) The changes introduced by this Service Bulletin are as follows:

(a) The existing bolts which install the brackets including the bracket PN 5W8317 to the flange FA1, have been changed from PN 4W0164 to PN 4W0165 to increase the bolt length.

(2) New bolts will be available for future replacement purposes.

D. Approval

The part number changes and/or part modifications described in Section 2 and 3 of this Service Bulletin have been shown to comply with the applicable Federal Aviation Regulations and are FAA-APPROVED for the engine model listed.

E. Compliance

Category Code 3.

Accomplish prior to revenue service.

F. Manpower

Estimated Manhours to incorporate the full intent of this Bulletin:

Venue	Estimated Manhours
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(1) In service:

(a) To gain access .. ..

(b) To embody .. ..

(c) To return engine to flyable  
status .. ..

Total: 33 mins

(2) At Overhaul: Not applicable.

G. Material - Price and Availability

(1) Modification kit is not required. Parts supplied as single line items.

(2) See "Material Information" section for prices and availability of future spares.

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## H. Tooling - Price and Availability

Special tools are not required to accomplish this Service Bulletin.

## I. Weight and Balance

- (1) Weight change .. .. . None
- (2) Moment arm .. .. . No effect
- (3) Datum .. .. . Engine front mount centreline  
(Powerplant Station (P.P.S.) 100)

## J. Electrical Load data

This Service Bulletin has no effect on the aircraft electrical load.

## K. References

- (1) Internal Reference No.  
  
EC88VJ593B
- (2) Other References  
  
Aircraft Maintenance Manual, 71-13-00, Maintenance Practices, and  
70-23-11, Torque Tightening Technique.

## L. Other Publications Affected

- (1) V2500 Power Plant Illustrated Parts Catalog, Chapter/Section 71-51-41.
- (2) V2500 Engine Illustrated Parts Catalog, Chapter/Section 71-51-41.



## 2. Accomplishment Instructions

### A. Prerequisite Instructions

- (1) Open the right fan cowl. (Refer to the Aircraft Maintenance Manual, 71-13-00, Maintenance Practices, TASK 71-13-00-010-010).

NOTE: For uninstalled engine (without the fan cowls), this step is not applicable.

### B. Rework Instructions

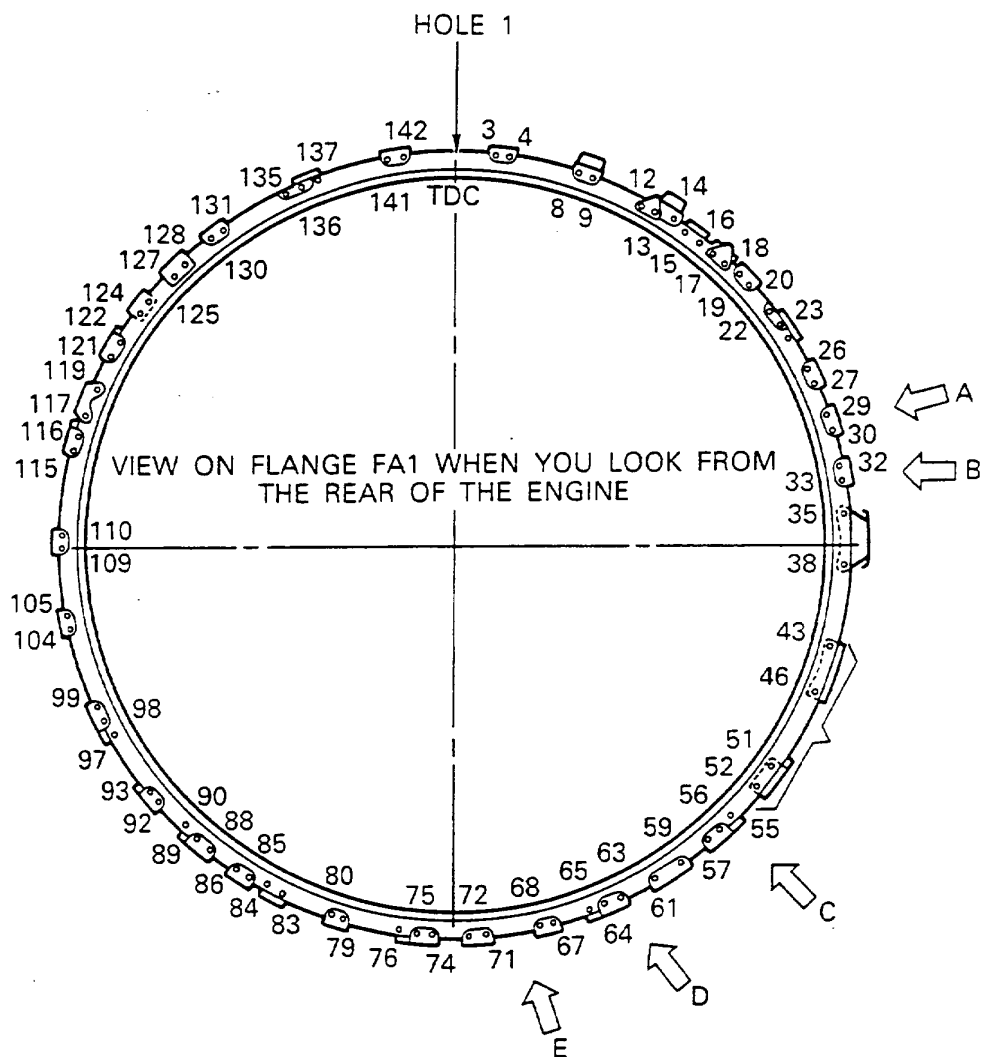
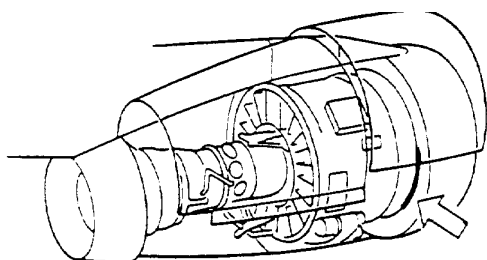
- (1) Find the flange holes 29, 30, 32, 33, 57, 63, 64, 67 and 68 on the flange FA1 where the brackets including bracket PN 5W8317 are installed (Refer to Figures 1 and 2.)
- (2) At each flange hole 29, 30, 32, 33, 57, 63, 64, 67 and 68, remove PN 4W0002, nut, and PN 4W0164, bolt from the flange FA1 and the brackets. (Refer to Figures 1 and 2.)
- (3) At each flange hole 29, 30, 32, 33, 57, 63, 67 and 68, install PN 4W0165, bolt, and PN 4W0002, nut to the brackets and the flange FA1. (Refer to Figures 1 and 2.)
- (4) At each flange hole 29, 30, 32, 33, 57, 63, 64, 67 and 68, tighten PN 4W0002, nut on PN 4W0165, bolt with a torque of 85-95 lbfin (9,6 - 10,7 Nm). (Refer to Figures 1 and 2, and the Aircraft Maintenance manual, 70-23-11, Torque Tightening Technique).

### C. Postrequisite Instructions

- (1) Close the right fan cowl. (Refer to the Aircraft Maintenance manual, 71-13-00, Maintenance Practices, TASK 71-13-00-410-010.)

### D. Recording Instructions

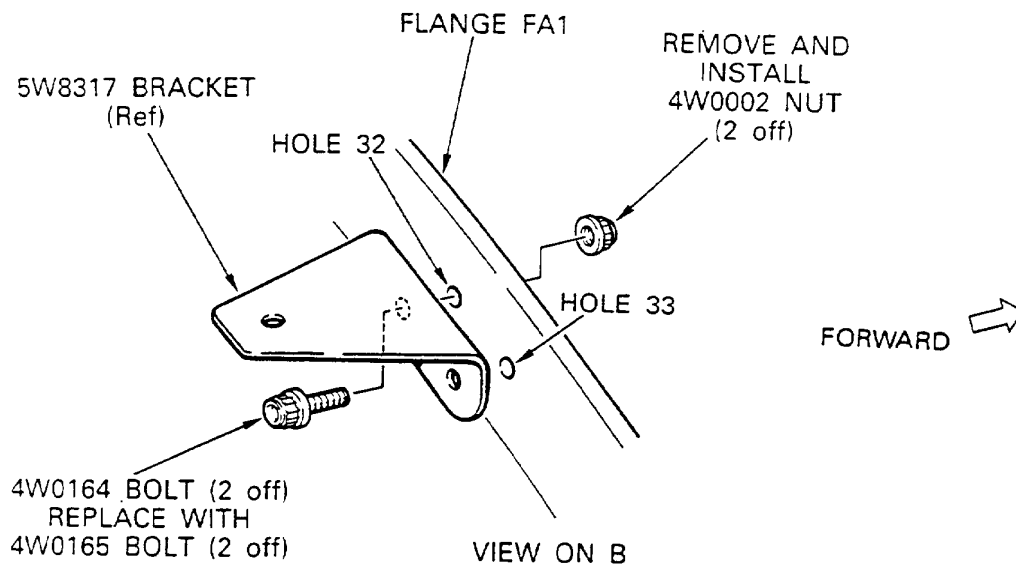
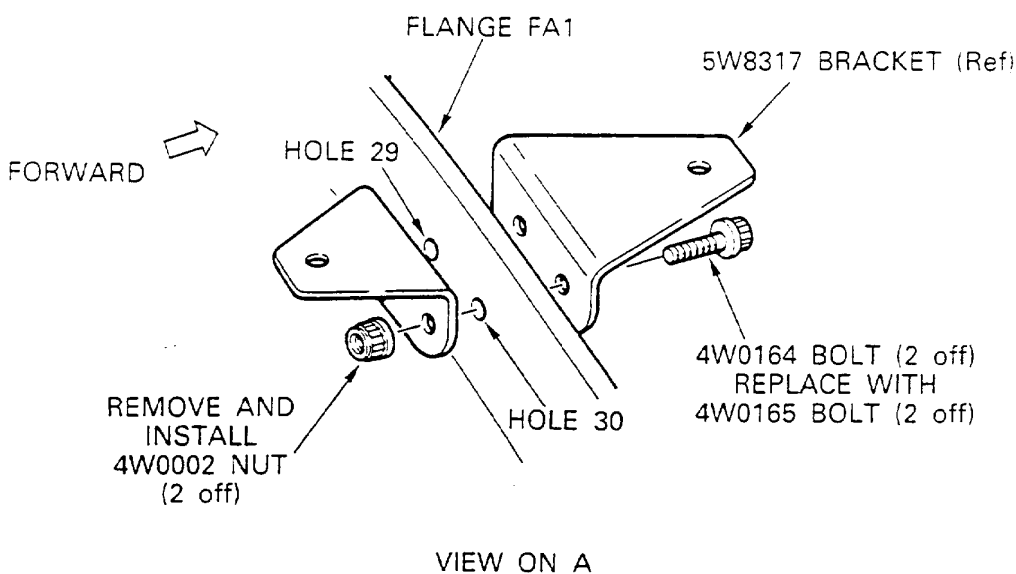
- (1) A record of accomplishment is necessary.



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Location of the flange holes at the flange FA1  
Fig.1

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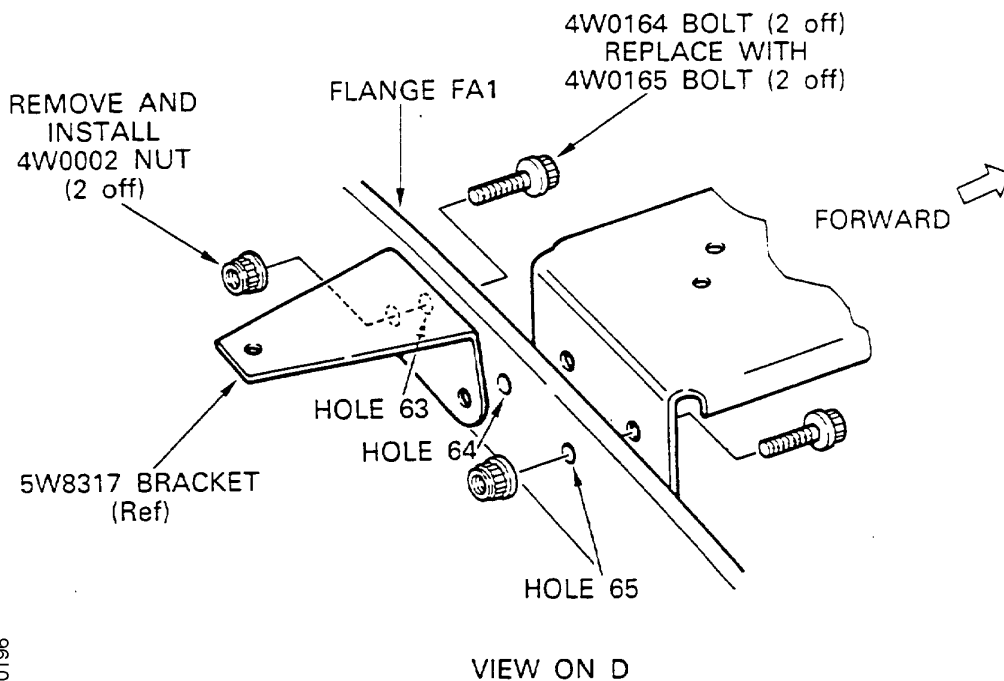
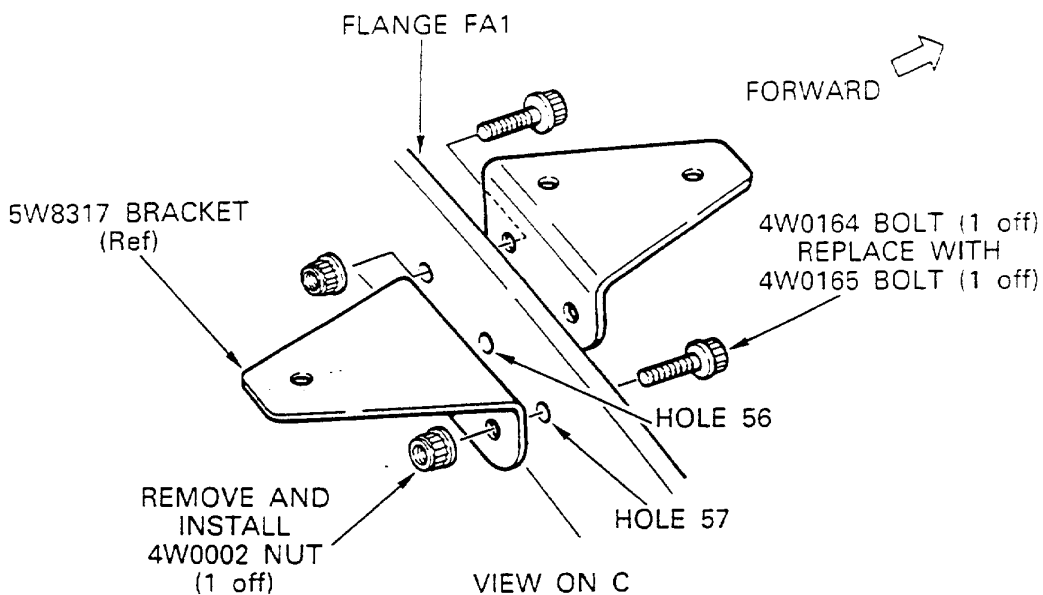
Replacement of the bracket attaching bolts on flange FA1  
Fig.2

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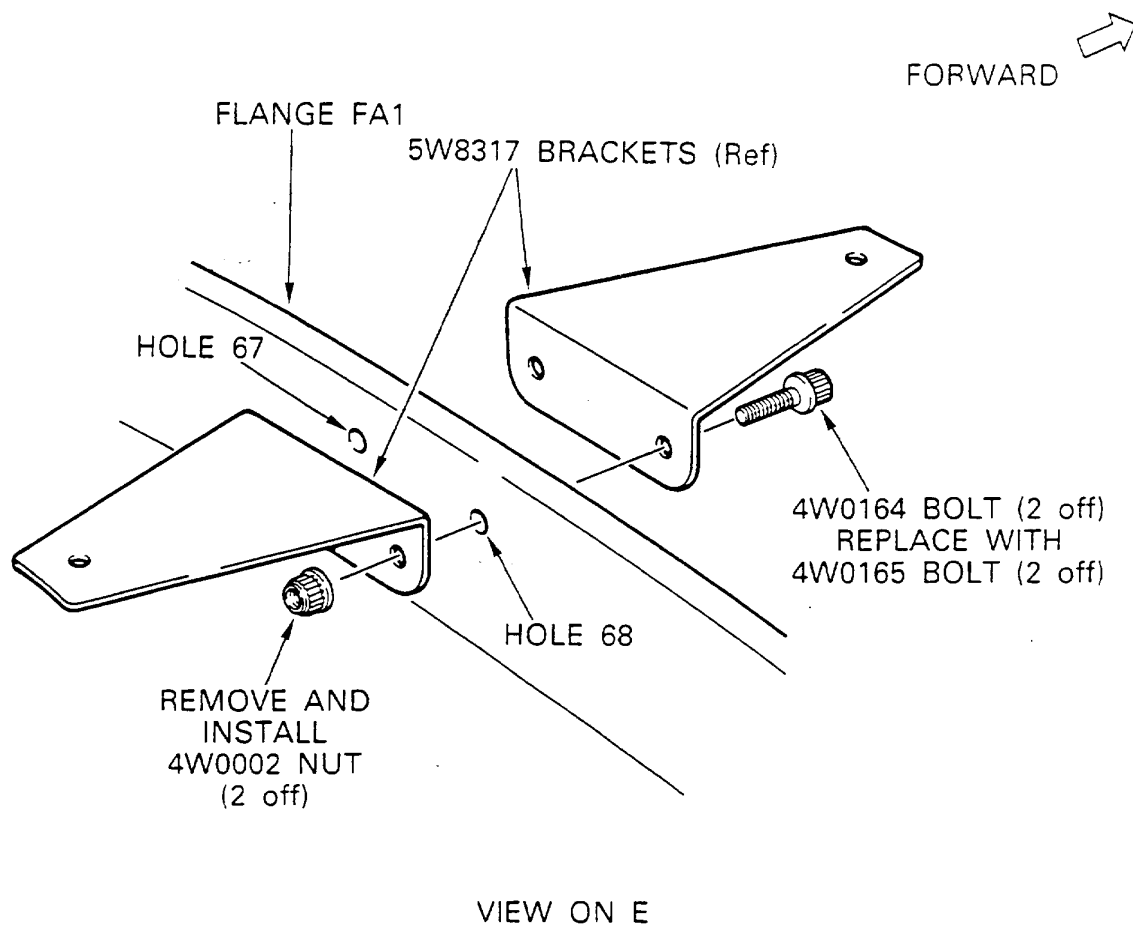
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Replacement of the bracket attaching bolts on flange FA1  
Fig.2

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Replacement of the bracket attaching bolts on flange FA1  
Fig.2

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3. Material Information

New Part No. (ATA No.)	Qty	Est'd Unit Price (\$)	Keyword	Old Part No. (IPC No.)	Instructions Disposition
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Applicability: For each V2500 Engine to incorporate this Bulletin.

A. Kits associated with this Bulletin:

None

B. Parts affected by this Bulletin:

4W0165	9	3.50	Bolt Machine Double Hex	4W0164	(A) (B) (1D)
(71-51-41)				(05-894)	

C. Instruction/Disposition Code Statements:

(A) New part is currently available for sale.

(B) Old part will continue to be available for sale on the other applications.

(1D) Old part can be used up on the other applications.

NOTE; The Estimated 1989 Unit Prices shown are provided for planning purposes only and do not constitute a firm quotation. Contact IAE's Spare Parts Sales Department for information concerning firm prices.

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