

ENGINE - ADDITION OF WASHERS FOR CONNECTION OF THE NF TERMINAL BLOCK AND THE EEC  
HARNESS-CORE - CATEGORY CODE 3 - MOD.ENG-72-0023

1. Planning Information

A. Effectivity

- (1) Aircraft: Airbus A320
- (2) Engine: V2500-A1 Engines, Serial No.V0014 through V0029

B. Reason

(1) Condition

Contact error may occur between the NF terminal block and NF and trim balance wiring harness.

(2) Background

Existing terminal studs on the NF terminal block have excessive shank length. Therefore the terminal of the EEC harness-core has play on the stud, even after the nut is fully screwed. Additional washer was introduced in order to fill the clearance between the nut and the terminal.

(3) Objective

Add a washer to each of six terminal studs to get a faultless contact.

(4) Substantiation

Substantiation test is not required.

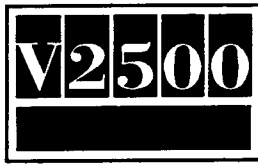
(5) Effects of Bulletin on the following shop functions:

Removal/Installation	Affected (See supplemental information)
Disassembly/Assembly	Not affected
Cleaning	Not affected
Inspection/Check	Not affected
Repair	Not affected
Testing	Not affected

(6) Supplemental Information

The Post-Service Bulletin configuration requires revision of the Engine Manual under Chapter 72-00-32 Removal and Installation, to show requirement for removing/installing six additional washers.

V2500-ENG-72-0023

**C. Description**

(1) The changes introduced by this Service Bulletin are as follows:

- (a) A washer, PN AN960C8 is added to each stud of the six studs on the NF terminal block. The washer places between the terminal of the EEC harness-core and the nut.
- (b) Fig/item numbers of the nuts to retain the terminals are also revised to be consistent with the call outs specified in the Engine Illustrated Parts Catalog.

**D. Approval**

The part number changes and/or part modifications described in Section 2 and 3 of this Service Bulletin have been shown to comply with the applicable Federal Aviation Regulations and are FAA-APPROVED for the Engine Model listed.

**E. Compliance**

Category Code 3

Accomplish prior to revenue service.

**F. Manpower**

Estimated Manhours to incorporate the full intent of this Bulletin:

Venue	Estimated Manhours
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(1) In service

(a) To gain access .. .. .

(b) To embody .. .. .

(c) To return engine to flyable status ..

TOTAL	26 minutes
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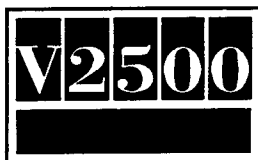
**G. Material - Price and Availability**

Modification kit not required. Parts supplied as single line item.

**H. Tooling - Price and Availability**

Special tools are not required.

V2500-ENG-72-0023



# SERVICE BULLETIN

## I. Weight and Balance

- (1) Weight change .. .. . None
- (2) Moment arm .. .. . No effect
- (3) Datum .. .. . Engine front mount centerline (Powerplant Section (P.P.S.100))

## J. Electrical Load Data

This Service Bulletin has no effect on the aircraft electrical load.

## K. References

- (1) Internal Reference No.  
EC88VJ709
- (2) Other References  
Aircraft Maintenance Manual.  
Standard Practices/Processes Manual.

## L. Other Publications Affected

- (1) V2500 Engine Illustrated Parts Catalog, 71-52-43.
- (2) V2500 Engine Manual, 72-00-40, Removal-02 and 72-00-40, Installation-09.

V2500-ENG-72-0023



## SERVICE BULLETIN

2. Accomplishment Instructions

## A. Prerequisite Instructions

## Procedure

- (1) Open the right fan cowl door
- (2) Open the right thrust reverser half.

## Supplementary Information

Refer to the Aircraft Maintenance Manual, 71-13-00, Maintenance Practices, TASK 71-13-00-010-010.

Refer to the Aircraft Maintenance Manual, 78-32-00, Maintenance Practices, TASK 78-32-00-010-010.

## B. Fitment Instructions

## Procedure

- (1) Remove two AS27831 Nuts from studs, position seven and eight.
- (2) Install an AN960C8 Washer to each of two studs nut removed.
- (3) Install an AS27831 Nut to each stud.
- (4) Torque the nuts to 10 to 11.5 lbfin (1,13 to 1,30 Nm).
- (5) Remove four AS27382 Nuts from studs, position one, two, five and six.
- (6) Install an AN960C8 Washer to each of four studs nut removed.
- (7) Install an AS27832 Nut to each stud.
- (8) Torque the nuts to 20 to 22 lbfin (2,26 to 2,48 Nm).

## Supplementary Information

Refer to Figure 1.

Refer to Standard Practices/ Processes Manual, 70-41-02, Torque Thread Values, TASK 70-41-02-400-501.

Refer to Standard Practices/ Processes Manual, 70-41-02, Torque Thread Values, TASK 70-41-02-400-501.

## C. Postrequisite Instructions

V2500-ENG-72-0023



## SERVICE BULLETIN

### Procedure

- (1) Do the Test No.5, EEC System Static Test.
- (2) Close the right thrust reverser half.
- (3) Close the right fan cowl door.

### Supplementary Information

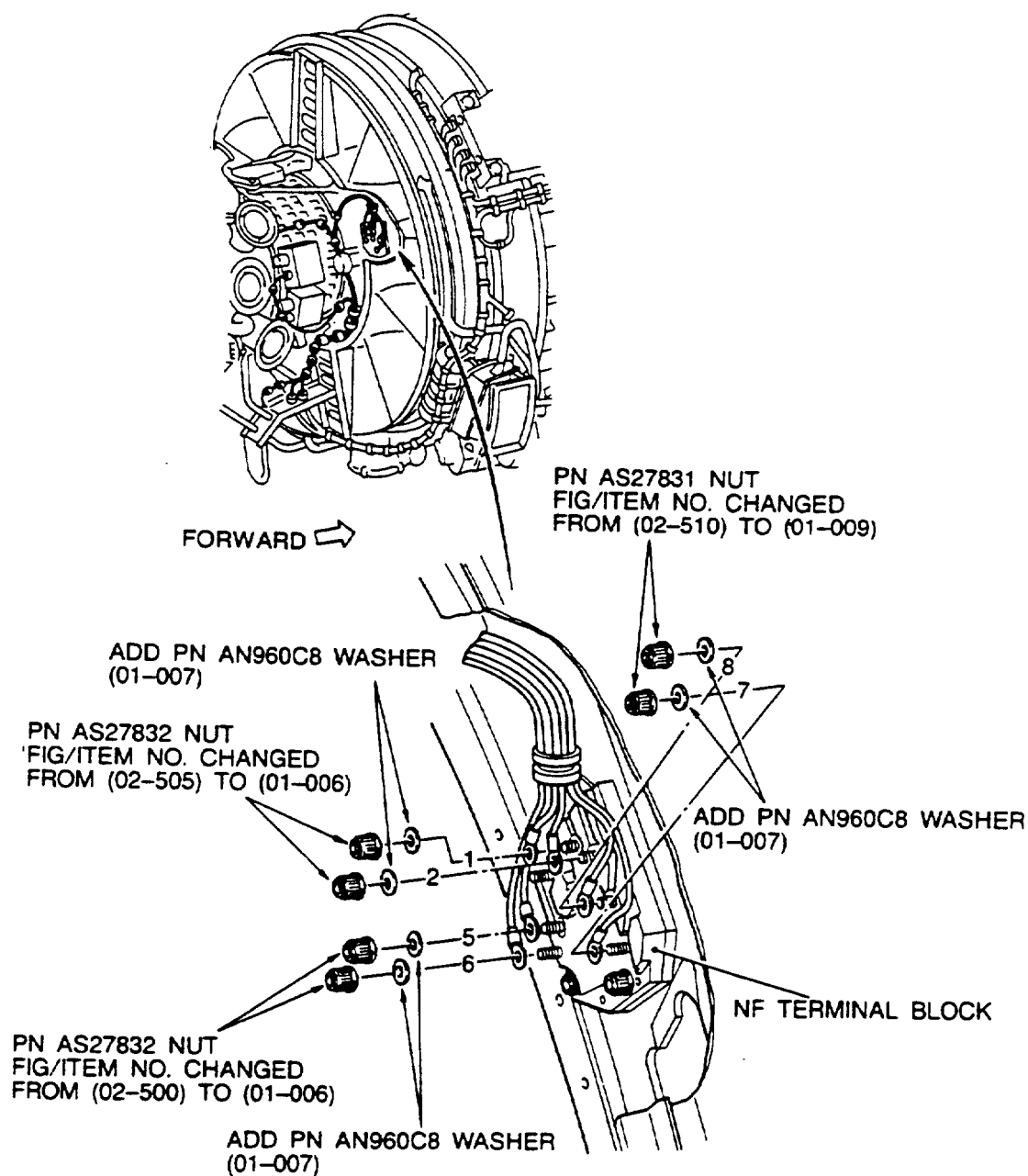
Refer to the Aircraft Maintenance Manual, 71-00-00, Adjustment/Test, TASK 71-00-00-700-019.

Refer to the Aircraft Maintenance Manual, 78-32-00, Maintenance Practices, TASK 78-32-00-410-010.

Refer to the Aircraft Maintenance Manual, 71-13-00, Maintenance Practices, TASK 71-13-00-410-010.

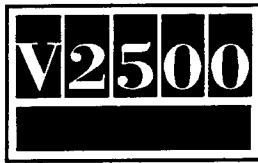
### D. Recording Instruction

- (1) A record of accomplishment is necessary.



Addition of the washers  
Fig.1

V2500-ENG-72-0023

3. Material Information

New Part No. (ATA No.)	Qty	Est'd Unit Price (\$)	Keyword	Old Part No. (IPC No.)	Instructions Disposition
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Applicability: For each V2500 Engine to incorporate this Bulletin.

A. Kits associated with this Bulletin:

None

B. Parts affected by this Bulletin:

AN960C8	6	0.03	Washer		(A)
(71-52-43)				(01-007)	

C. Instruction/Disposition Code Statements:

(A) New part is currently available for sale.

V2500-ENG-72-0023

