



ENGINE - INCORPORATION OF FIRE SHIELDS ON THE NO.1 STRUT FAIRING PANEL A/O - CATEGORY
CODE 3 - MOD.ENG-72-0024

1. Planning Information

A. Effectivity

- (1) Aircraft: Airbus A320
- (2) Engine: V2500-A1 Engines. Serial No.'s V0058, V0064 and V0065

B. Reason

(1) Condition

It was determined during certification testing that the existing No.1 Strut Fairing Panel Assembly may not provide sufficient protection for the Thrust Reverser Firewall Seal.

(2) Background

This condition was noted during certification testing.

(3) Objective

To incorporate the Fire Shields on the No.1 Strut Fairing Panel A/O to provide protection for the Thrust Reverser Firewall Seal.

(4) Substantiation

Substantiation is not required.

(5) Effects of Bulletin on Workshop Procedures:

Removal/Installation - Not affected.
Disassembly/Assembly - Affected (See Supplemental Information).
Cleaning - Not affected.
Inspection/Check - Not affected.
Repair - Not affected.
Testing - Not affected.

(6) Supplemental Information

- (a) The disassembly/assembly of the Post-Service Bulletin configuration requires instructions for the Fire Shields and their attaching parts.

C. Description

- (1) The changes introduced by this Bulletin is as follows:

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- (a) The Fire Shields and their attaching parts are being added to the No.1 Strut Fairing Panel (A/0), to provide the thermal protection for the Thrust Reverser Firewall Seal. (See Figure 2).
- (b) In agreement with the above, the No.1 Strut Fairing Panel (A/0) is reworked to include drilled holes which permit the installation of the Fire Shields on the Fairing Panel with the Screws, Nuts, and Washers. (See Figure 2).
- (2) The existing No.1 Strut Fairing Panel (A/0) can be reworked to a new configuration. (See Figure 2).
- (3) New No.1 Strut Fairing Panel (A/0) and Fire Shields including their attaching Screws, Nuts, and Washers will be available for future replacement purposes.

D. Approval

The part number changes and/or part modifications described in Sections 2 and 3 of this Service Bulletin have been shown to comply with the applicable Federal Aviation Regulations and are FAA-APPROVED for the Engine Model listed.

E. Compliance

Category Code 3.

Accomplish prior to revenue service.

F. Manpower

Estimated manhours to incorporate the full intent of this Bulletin:

Venue	Estimated Manhours
(1) In service	
(a) To gain access	
(b) To embody	
(c) To return engine to flyable status	
	Total: 1 hr 24 mins
(2) At overhaul	Not applicable

G. Material – Price and Availability

- (1) Modification Kit not required. Parts supplied as single line items.

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- (2) See "Material Information" section for prices and availability of future spares.

H. Tooling - Price and Availability

Special tools are not required to accomplish this Service Bulletin.

I. Weight and Balance

- | | | | | | |
|-----|---------------|----|----|----|--|
| (1) | Weight Change | .. | .. | .. | None |
| (2) | Moment Arm | .. | .. | .. | No effect |
| (3) | Datum | .. | .. | .. | Engine front mount centerline
(Powerplant Station (P.P.S.) 100) |

J. Electrical Load Data

This Service Bulletin has no effect on the aircraft electrical load.

K. References

- (1) Internal Reference No.

EC88VN045B

- (2) Other References

V2500 Engine Illustrated Parts Catalog, Chapter/Section 72-32-03.

A320 Aircraft Maintenance Manual, 78-32-00, Maintenance Practices, and 78-32-01, 78-32-02, 78-32-03, 78-32-04, Removal/Installation, and 70-23-11, Torque Tightening Technique.

V2500 Standard Practices/Processes Manual, 70-09-00, Marking of Parts.

V2500 Overhaul Processes and Consumables Index.

L. Other Publications Affected

- (1) V2500 Engine Illustrated Parts Catalog, Chapter/Section 72-32-03.
- (2) V2500 Engine Manual, 72-32-00, LP Compressor/Intermediate Case Module - Disassembly, TASK 72-32-00-030-002, Assembly, TASK 72-32-00-430-014 and 72-32-03, Intermediate Structure - Cleaning, TASK 72-32-03-100-002, Inspection/Check, TASK 72-32-03-200-002.

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2. Accomplishment Instructions

A. Pre-Requisite Instructions

- (1) Remove the Thrust Reverser Halves. (Refer to 1.K.(2), Aircraft Maintenance Manual, 78-32-00, Maintenance Practices, TASK 78-32-00-010-010, and Removal/Installation, TASK 78-32-01-000-010 and 78-32-03-000-010, or TASK 78-32-02-000-010 and TASK 78-32-04-000-010, as applicable).

NOTE: This step is not applicable if the Thrust Reverser Halves have been removed from the aircraft and there is access to the parts.

B. Rework Instruction

- (1) Find 5A2183, No.1 Strut Fairing Panel A/O on the Fan Frame No.1 Strut. (Refer to Figure 1).
- (2) Place 740-0500-501, Fire Shield over existing the Fairing Panel attaching Bolt on the inner left side of the Fairing Panel. Butt the Fire Shield with the top of the Fairing Panel and maintain a vertical gap of 0.13in. (3,3 mm.) between the Fairing Panel and the Fire Shield. Hold the Fire Shield with a clamp in position. (Refer to Figure 2, Sheet 1).
- (3) Drill two 0.130 to 0.137in. (3,30 to 3.48 mm.) diameter holes from two pilot holes in the Fire Shield through the Fire Shield and the Fairing Panel. Remove the clamp, and keep the Fire Shield. (Refer to Figure 2, Sheet 1).
- (4) Remove all burrs from the holes in the Fire Shield and the Fairing Panel. (Refer to Figure 2, Sheet 1).
- (5) Remove only enough Bolts, Nuts, Washers and Strut Corner Fillers to peel back the Fairing Panel allowing access as follows: (Refer to Figure 2, Sheet 2).
 - (a) Remove the three 4W0002, Nuts, 4W0166, Bolts, the six AN960C416, Washers, the four 4W0106, Bolts, and AN960C10, Washers.
 - (b) Remove the two 4W0104, Bolts, the two 4W2621, Washers, and 5A2126, LH Panel Corner Filler.
 - (c) Remove the two 4W0104, Bolts, the two 4W2621, Washers, and 5A2127, RH Panel Corner Filler.
 - (d) Move the Fairing Panel rearwards to get access to the fore side of Fairing Panel.

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- (6) Attach 740-0500-501, Fire Shield to the Fairing Panel with two NAS1802-04-6, Screws, AN960C4L, Washers, and MS21043-04, Nuts. Torque the Nuts to 6 – 7 lbfin (0,68 – 0,80 Nm). (Refer to Figure 2, Sheet 1 and 1.K.(2), Aircraft Maintenance Manual, 70-23-11, Torque Tightening Technique).
- (7) Do steps B.(1), B.(2), B.(3), B.(4) and B.(6) again to install 740-0500-502, Fire Shield, on the inner right side of the Fairing Panel.
- (8) Put the Fairing Panel back to the Fan Frame and the Fan Case, and install the removed Bolts, Nuts, Washers and the Strut Corner Fillers as follows: (Refer to Figure 2, Sheet 2, and 1.K.(2), Aircraft Maintenance Manual, 70-23-11, Torque Tightening Technique, and 1.K.(4)).
- (a) Install the three 4W0166, Bolts, the six AN960C416, Washers, and the three 4W0002, Nuts, to safety the Fairing Panel and the Fan Case Bracket. Torque the Bolts to 85 – 100 lbfin (9,60 – 10,50 Nm).
 - (b) Install the four AN960C10, Washers and 4W0106, Bolts, to safety the Fairing Panel and the Fan Frame. Torque the Bolts to 36 – 40 lbfin (4,07 – 4,52 Nm).
 - (c) Remove the sealant on the mating surfaces of the two Panel Corner Fillers and the Fairing Panel Corners.
 - (d) Clean the mating surfaces of the two Panel Corner Fillers with a clean cloth or a soft brush, and CoMat 01-076, Methylethylketone.
 - (e) Apply thin layer of CoMat 08-014, Primer to the mating surfaces of the two Panel Corner Fillers. Air dry for 30 minutes minimum.
- NOTE: Apply the primer in less then eight hours after the mating faces are cleaned.
- (f) Apply CoMat 08-045, Sealant (RTV-103), to the mating surfaces of the two Panel Corner Fillers.
 - (g) Install 5A2127, RH Panel Corner Filler, with the two 4W2621, Washers, and the two 4W0104, Bolts. Torque the two Bolts to 36 – 40 lbfin (4,07 – 4,52 Nm).
 - (h) Install 5A2126, LH Panel Corner Filler, with the two 4W2621, Washers, and the two 4W0104, Bolts. Torque the two Bolts to 36 – 40 lbfin (4,07 – 4,52 Nm).
- (9) Renumber by the Electrolytic etch adjacent to the existing Part Number on the Fairing Panel as follows. (Refer to Figure 2, Sheet 1, and 1.K. (3)).



Existing	Renumber
5A2183	5A0212

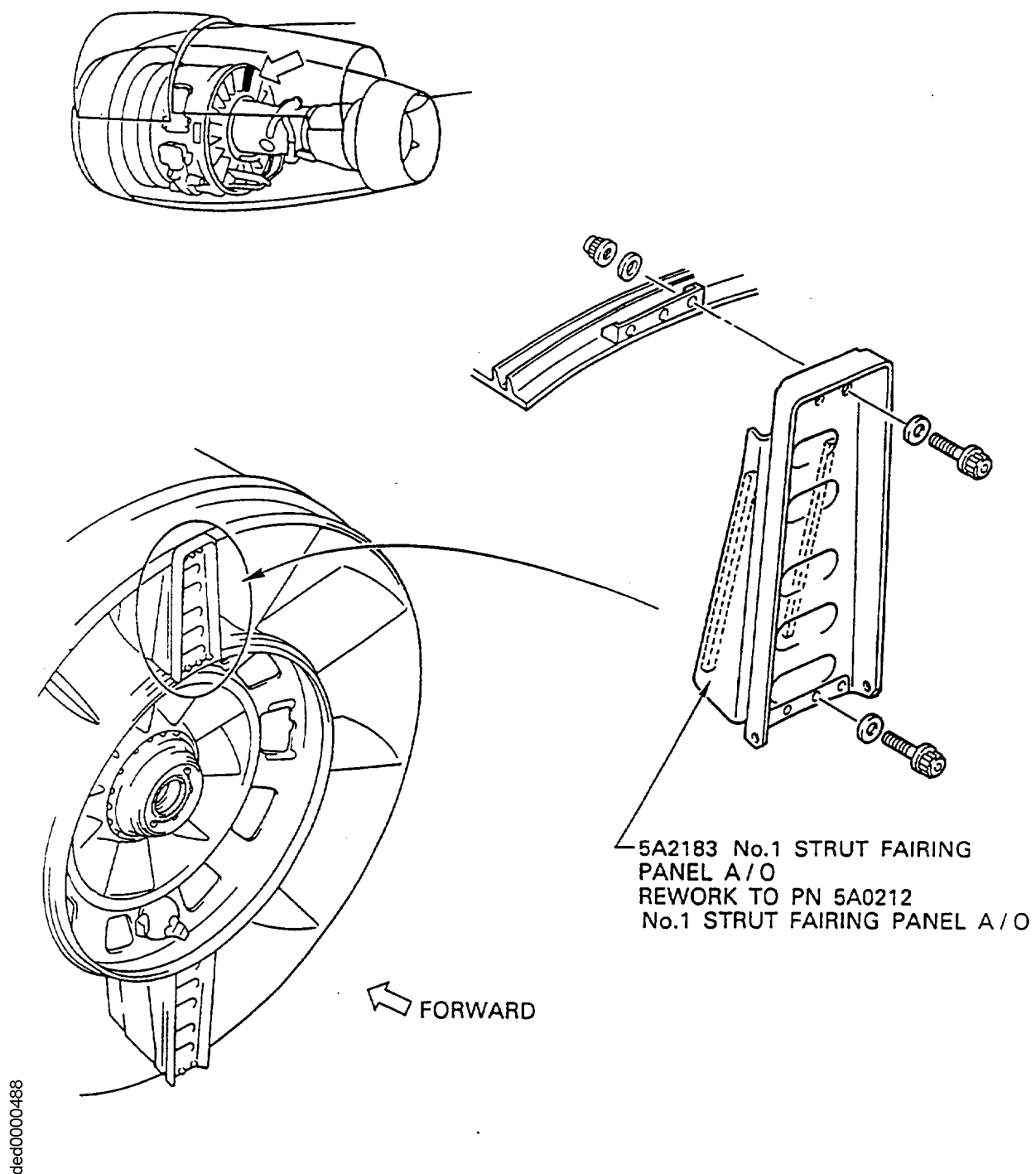
C. Post-Requisite Instructions

- (1) Install the Thrust Reverser Halves. (Refer to 1.K.(2), Aircraft Maintenance Manual, 78-32-00, Maintenance Practices, TASK 78-32-00-010-010, and Removal/Installation, TASK 78-32-01-400-010 and TASK 78-32-03-400-010 or TASK 78-32-02-400-010 and TASK 78-32-04-400-010 as applicable).

NOTE: This step is not applicable if the Thrust Reverser Halves have been removed from the aircraft and there is access to the parts.

D. Recording Instructions

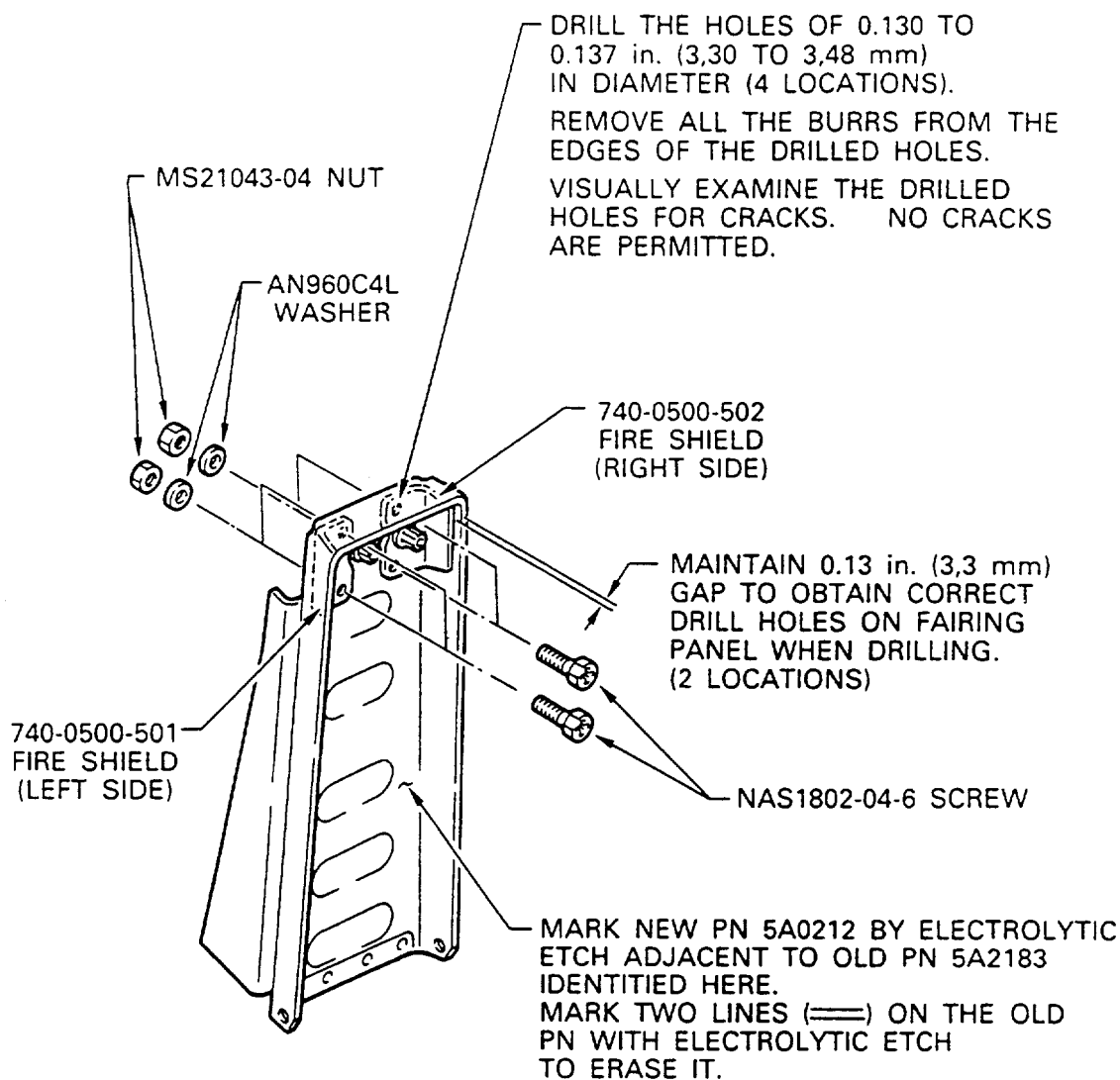
- (1) A record of accomplishment is necessary.



Location of the No.1 Strut Fairing Panel A/O
Fig.1



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← FORWARD

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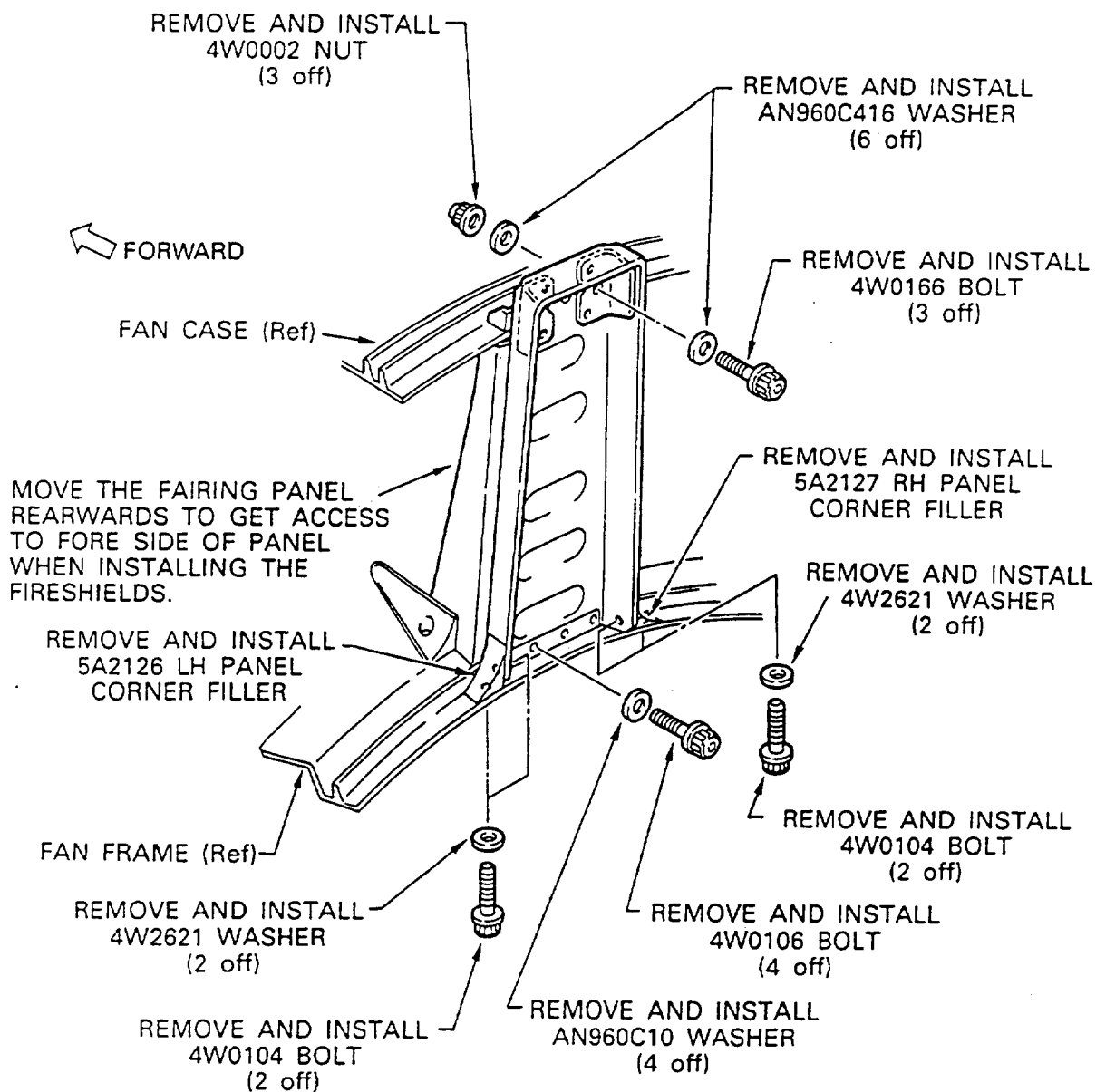
Rework of the No.1 Strut Fairing Panel A/0
Fig.2, Sheet 1 of 2

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Rework of the No.1 Strut Fairing Panel A/O
Fig.2, Sheet 2 of 2

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3. Material Information

New Part No. (ATA No.)	Qty	Est'd Unit Price (\$)	Keyword	Old Part No. (IPC No.)	Instructions Disposition
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Applicability: For each V2500 Engine to incorporate this Bulletin.

A. Kits associated with this Bulletin:

None

B. Parts affected by this Bulletin:

New Part No. (ATA No.)	Qty	Est'd Unit Price (\$)	Keyword	Old Part No. (IPC No.)	Instructions Disposition
5A0212 (72-32-03)	1	1234.00	Panel A/O Fairing	5A2183 (02-300)	(A)(B)(S1) (1D)
740-0500- 501 (72-32-03)	1	-	Shield, Fire	--- (02-400)	(A)(C)(S1) (S2)
MS21043-04 (72-32-03)	2	-	Nut	--- (02-405)	(A)(D)(S1) (S2)
AN960C4L (72-32-03)	2	-	Washer	--- (02-407)	(A)(D)(S1) (S2)
NAS1802-04 -6 (72-32-03)	2	-	Screw	--- (02-410)	(A)(D)(S1) (S2)
740-0500- 502 (72-32-03)	1		Shield, Fire	--- (02-420)	(A)(C)(S1) (S2)
MS21043-04 (72-32-03)	2	-	Nut	--- (02-425)	(A)(D)(S1) (S2)
AN960C4L (72-32-03)	2	-	Washer	--- (02-427)	(A)(D)(S1) (S2)
NAS1802-04 -6 (72-32-03)	2	-	Screw	--- (02-430)	(A)(D)(S1) (S2)

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C. Instruction/Disposition Code Statements:

- (A) New part is currently available for sale.
- (B) Old part will no longer be available for sale.
- (C) Part coded (C) may be obtained from:

ROHR INDUSTRIES, INC.
FOOT OF H STREET P.O. BOX 878
CHULA VISTA, CA 92012

- (D) Parts coded (D) are standard parts and should be obtained from standard hardware supplier.
- (S1) New Parts Coded (S1) must be fitted as a set.
- (S2) Additional item.
- (1D) Rework Old Part Number and reidentify to the New Part Number.

NOTE: The estimated 1990 unit prices shown are provided for planning purposes only and do not constitute a firm quotation. Consult the IAE Price Catalog or contact IAE's Spare Parts Sales Department for information concerning firm prices.

