

ENGINE - REWORK THE BIFURCATION DISCONNECT PANEL ASSEMBLY TO EXTEND SEALING FEATURES -
CATEGORY CODE 4 - MOD.ENG-72-0062

Printed in Great Britain

1. Planning Information

A. Effectivity

(1) Aircraft: Airbus A320

(2) Engines: V2500-A1 Engines prior to V0096 excluding V0090 and V0092

B. Reason

Bifurcation disconnect panels on existing engines have two relief holes for manufacturing purposes only. These holes are required to be filled with cold curing silicone rubber compound.

C. Compliance

CATEGORY CODE 4

Accomplish at the first visit of an engine or module to a maintenance base capable of compliance with the accomplishment instructions regardless of the planned maintenance action or the reason for engine removal.

D. Approval

The "Compliance" statement and the requirement for accomplishing this modification specified in Paragraph F. of the Service Bulletin has been shown to comply with the applicable Federal Aviation Regulations and are FAA-APPROVED for the single model listed.

E. References

(1) Internal Reference No.

89VR858

(2) Other references

A320 Aircraft Maintenance Manual.

Standard Practices Manual.

F. Action

(1) Place warning notice not to start engine SUBTASK 78-32-99-941-400.

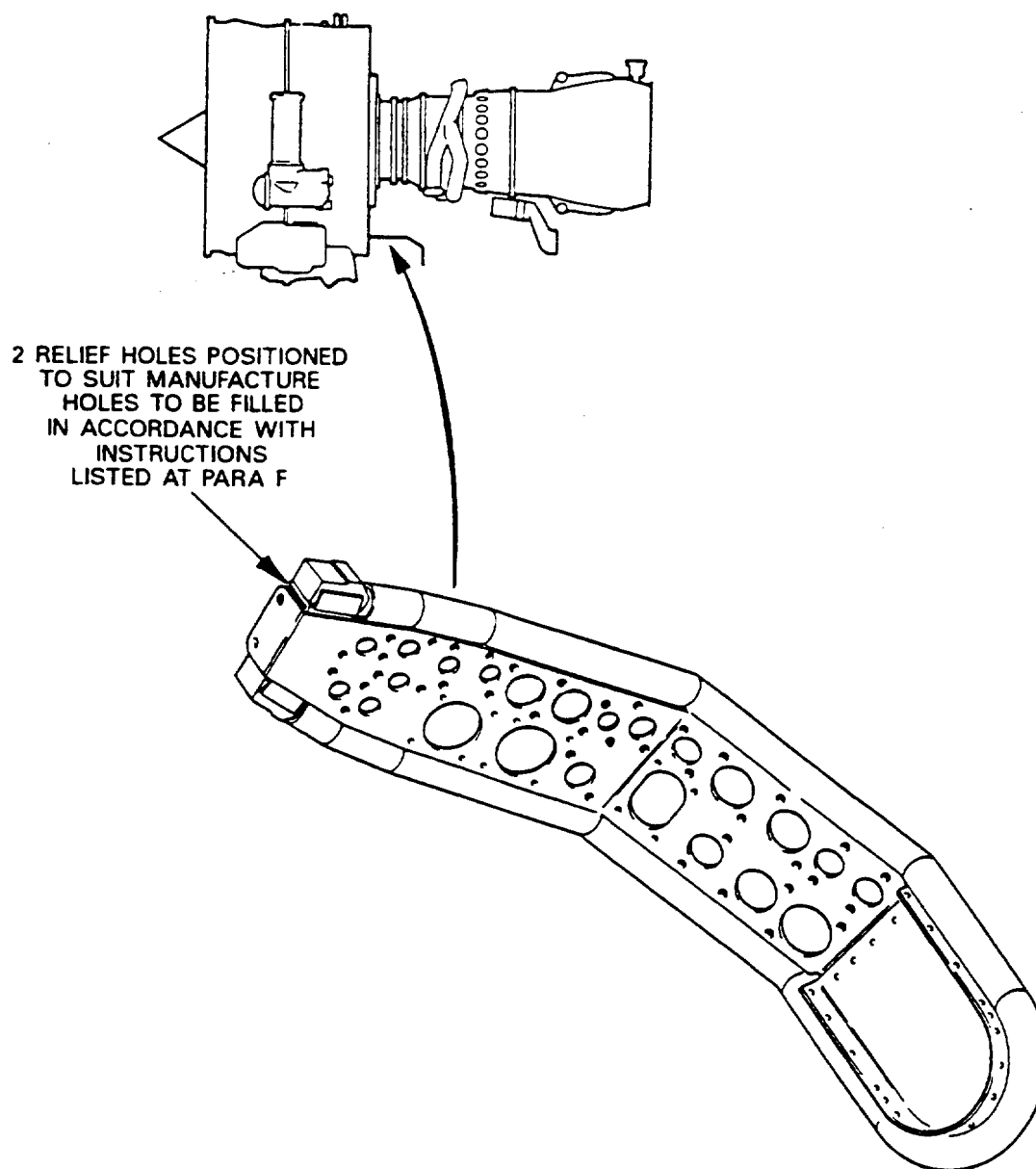
(2) Allow 5 minutes elapsed time from shutdown SUBTASK 78-32-00-869-400.

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- (3) Place warning notice on cockpit panel SUBTASK 78-32-00-869-401.
- (4) Open the cowl panel SUBTASK 78-32-00-010-404.
- (5) De-activate hydraulic control unit SUBTASK 78-32-00-040-403.
- (6) Open thrust reverser half SUBTASK 78-32-00-010-407.
- (7) Locate Bifurcated disconnect panel 6A3361 (See fig.1).
- (8) Apply cold curing Silicone compound CoMat 08-013 to the two manufacturing holes in accordance with:

SUBTASK 70-36-02-360-020. TOTAL: 30 mins.
- (9) Close thrust reverser half SUBTASK 78-32-00-440-403.
- (10) Activate hydraulic control unit SUBTASK 78-32-00-440-403.
- (11) Close cowl panel SUBTASK 78-32-00-410-408
- (12) Remove warning notices SUBTASK 78-32-00-942-400.



Bifurcation Panel
Fig.1

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