International Aero Engines

SERVICE BULLETIN

ENGINE - FUEL AND CONTROL - ENGINE DEDICATED ALTERNATOR (EDA) STATOR UNIT - ELECTRICAL RECEPTACLE INSPECTION AND CONNECTOR TIGHTENING PROCEDURE - CATEGORY CODE 3 - MOD.ENG-73-0034

1. Planning Information

A. Effectivity

(1) Aircraft: Airbus A320
McDonnell Douglas MD90

(2) Engine: V2500-A1 Engines prior to Serial No.V0232 (See Note)
V2500-D5 Engines prior to Serial No.V20069

NOTE: No repeat action is necessary where accomplishment has previously been completed.

B. Reason

Incorrect assembly of the harness connectors and the EDA stator receptacles may result in contact burning. This reissue accommodates later engine serial numbers that require to be checked.

C. Compliance

Category Code 3

Accomplish the intent of this Service Bulletin at the first 'A' check after receipt of this bulletin.

D. Approval

The 'compliance' statement and the procedure described in paragraph F of this Service Bulletin have been shown to comply with the applicable Federal Aviation Regulations and are FAA-APPROVED for the Engine Model listed.

E. References

(1) Internal Reference No.

ADR90909

EC91VR735

(2) Other references

The A320/V2500 Aircraft Maintenance Manual.

The V2500 Component Maintenance Manual, Electrical Harnesses and Cables.

F. Action

(1) Remove the electrical harness connectors from the EDA stator receptacles.

Refer to the Aircraft Maintenance Manual, Chapter 73-22-38.

(2) Inspect the stator receptacles.

(a) Carry out a visual examination of the receptacle pins.

(i) If the pins are gold or grey the stator is acceptable.

(ii) It the pins are blackened with material loss less than 75 per cent of the total circumference of the pin and less than 25 per cent depth of the pin thickness, the mating sockets in the electrical harness connector must be replaced prior to the next flight.

The engine may then operate normally until the next 'A' Check when the stator and the harness sockets must be replaced.

(iii) If the pins are burned in excess of (ii), replace the stator and the mating electrical harness sockets. Refer to the V2500 Component Maintenance Manual, electrical Harnesses and Cables, Chapter 71-51-41, VRS1035.

(3) If required install the EDA stator unit as instructed in the Aircraft Maintenance Manual, Chapter 73-22-38.

(4) Tighten the stator electrical connectors.

(a) Tighten the connector by hand until the red 'mating band' on the receptacle is covered by the connector shell.

(b) Using only a 'C' spanner to fit 3/4 in. to 2.in. diameter:

(i) Engage the dog of the 'C' spanner in one of four slots on the connector shell.

(ii) Using the spanner, turn the connector shell until resistance is felt and then 1/16th turn further.

(iii) Apply a torque stripe to the connector and receptacle to reference the connector position. Refer to Standard Practices Manual SUBTASK70-09-00-400-002. Use CoMat 06-160.

G. A record of accomplishment is necessary.