V2500 International Aero Engines

V2500 Propulsion System — Nacelle

SERVICE BULLETIN

Date: December 11, 2002

Subject: Transmittal of Revision 3 to Service Bulletin Number V2500-NAC-71-0259

Service Bulletin Revision History:

Event	Date
Original Issue	December 10, 1999
Revision 2	November 8, 2000
Revision 3	December 11, 2002

Reasons for issuance of Revision:

(1) To add re-identification instructions to the accomplishment section of the bulletin.

Effect on Past Compliance:

None.

List of Effective Pages:

Page No.	Rev No.	Date
1 – 21	3	December 11, 2002

NACELLE - POWERPLANT – AIR INLET COWL – INSTALLATION OF A FAN COWL DOOR HOLD OPEN DEVICE

MODEL APPLICATION

V2500-A1
V2522-A5
V2524-A5
V2527-A5
V2527E-A5
V2527M-A5
V2530-A5
V2533-A5

BULLETIN INDEX LOCATOR

71-00-00

COMPLIANCE CATEGORY CODE
4, 8

INTERNAL REFERENCE No
MM/SM 96VN208D/E/F

1. Planning Information

A. Effectivity

(1) Aircraft:

(a) Airbus A319
(b) Airbus A320
(c) Airbus A321

(2) Nacelle:

(a) All V2500-A1 Inlet Cowls.
(b) V2522-A5 Inlet Cowls prior to 1108001.
(c) V2524-A5 Inlet Cowls prior to 1108001.
(d) V2527-A5 Inlet Cowls prior to 1108001.
(e) V2527E-A5 Inlet Cowls prior to 1108001.
(f) V2530-A5 Inlet Cowls prior to 1108001.
(g) V2533-A5 Inlet Cowls prior to 1108001.

B. Concurrent Requirements

None.

C. Reason

(1) Condition

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

Several instances have been reported of Fan Cowl Doors not being correctly latched (as per manual requirements) prior to flight. This may result in actual Door loss in flight.

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

Several instances have been reported of the rubber block at each end of the Hold Open Device degrading or falling off.

(2) Background

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

There have been several reports from Operators of actual Fan Cowl Door loss due to the Doors not being fully latched. Several Service Bulletins (ref:

para 1.L) have been issued recently to address this problem. This Service Bulletin provides additional enhanced visibility of unlatched doors.

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

The Hold Open Device was introduced to enhance visibility of unlatched doors. However, the rubber Pads on the Hold Open Device have, in some instances, degraded or fallen off.

(3) Objectives

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

To maintain Engine reliability by the introduction of a Hold Open Device (H.O.D) to the Air Inlet Cowl Door Locator. The H.O.D prevents the Fan Cowl Doors from closing fully when not latched and produces a visible gap between the Fan Cowl Doors and the Air Inlet Cowl.

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

To give the Operator the option of reworking the Hold Open Device to the latest specification.

(4) Substantiation

This modification which has been the subject of a successful trial installation.

(5) Effect of Bulletin on:

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

(1)	Removal/Installation	Affected
(2)	Disassembly/Assembly	Affected
(3)	Cleaning	Not affected
(4)	Inspection/Check	Not affected
(5)	Repair	Not affected
(6)	Testing	Not affected

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

(1)	Removal/Installation	Not affected
(2)	Disassembly/Assembly	Not affected
(3)	Cleaning	Not affected
(4)	Inspection/Check	Not affected
(5)	Repair	Not affected
(6)	Testing	Not affected

(6) Supplemental Information

(a) The A320/V2500-A1, A319/A320/A321/V2500-A5 Aircraft Maintenance Manual, Chapters 71-13-00, 71-13-11 and 71-13-16 will be revised to include steps to instruct operation of the H.O.D.

(b) The A320/V2500 Engine Manual, Chapter 71-11-11 will be revised to include steps for the removal and installation of the new H.O.D.

D. Description

(1) The changes introduced by this Service Bulletin are as follows:

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

- Two Bolts that hold the Fan Cowl Door Guide Bar to the Air Inlet Cowl are removed.
- The new Hold Open Device is then installed to the Guide bar using two longer Bolts.
- A function check is then performed on the new Hold Open Device.

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

- The H.O.D is disassembled and removed from the Nacelle.
- The rubber block at each end of the Blocker Bar Assembly is removed and the Blocker Bar Assembly is reidentified.
- The H.O.D is installed.
- A function check is then performed on the new Hold Open Device.

E. Approval

The technical content of this Service Bulletin has been approved under the authority of the DGAC Design Organisation Approval No. F.JA.02.

F. Compliance

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

Category 4.

Accomplish at the first visit of the Nacelle or Nacelle component to a maintenance base capable of compliance with the Accomplishment Instructions regardless of the planned maintenance action for the Nacelle or Nacelle component.

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

Category 8.

Accomplish based upon experience with the prior configuration.

G. Manpower

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

Estimated man-hours to incorporate the intent of this Service Bulletin on each engine:

VENUE		ESTIMATED MAN-HOURS
(1)	At Overhaul	0.75 M/Hr
		Total 0.75 M/Hr

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

Estimated man-hours to incorporate the intent of this Service Bulletin on each engine:

VENUE		ESTIMATED MAN-HOURS
(1)	At Overhaul	0.5 M/Hr
		Total 0.5 M/Hr

NOTE: Manhours are provided for planning purposes only. No labor reimbursement is provided under the terms of this service bulletin offering.

H. Material Cost and Availability

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

The parts to accomplish this Service Bulletin are available from the manufacturer as Kit No V2571259-553

Operators with units listed in Paragraph 1.A should submit a purchase order for the applicable quantity of kits. The purchase order must specify this Service Bulletin number and the parts listed herein. Operators will have one year from the issue date of this Service Bulletin to place an order. After one year, kits will no longer be available and operators will have to order parts individually at catalog prices.

Direct request to:

Goodrich
850 Lagoon Drive
Chula Vista, CA 91910-2098 USA

Attn: Regional Business Manager – MZ107A
(Service Bulletin No. V2500-NAC-71-0259)

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

None.

I. Tooling - Cost and Availability

None.

J. Weight and Balance

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

(1)	Weight change.....	1.389 LB (0.63 Kg)
(2)	Moment arm.....	50.2 in (1275.1mm) Forward of Datum
(3)	Datum	Engine Front Mount Centreline (Powerplant Station PPS 100.00)

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

None.

K. Electrical Load Data

Not Affected.

L. References

Publication	Chapter/Section
A320/V2500-A1 A319/A320/A321/V2500-A5 Aircraft Maintenance Manual(M-V2500-1IA)	71-13-00
SPP-V2500-1IA Standard Practices/Processes Manual	70-09-00 70-41-00
IAE PCI-V2500-1IA Overhaul Processes And Consumable Index	Section II
Service Bulletins	V2500-NAC-71-0227 V2500-NAC-71-0235 V2500-NAC-71-0256
This SB is the subject of AIRBUS Aircraft Modification No 30869/P6925. AIRBUS Service Bulletin No. A320-71-1028	

M. Other Publications Affected

Publication	Chapter/Section
A320/V2500-A1 Engine Illustrated Parts Catalog	71-11-11
A319/A320/A321/V2500-A5 Engine Illustrated Parts Catalog – all variants	71-11-11
A320/V2500-A1 Powerplant Illustrated Parts Catalog	71-11-11
A319/A320/A321/V2500-A5 Powerplant Illustrated Parts Catalog– all variants	71-11-11
A320/V2500 Engine Manual (E-V2500-1IA)	71-11-11

2. Accomplishment Instructions

A. Pre-requisite Instructions

(1) Open the Fan Cowl Doors as instructed in the A320/V2500-A1, A319/A320/A321/V2500-A1/A5 Aircraft Maintenance Manual, Task 71-13-00-010-010.

B. Rework Instructions

Part A - For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

WARNING: IT IS THE RESPONSIBILITY OF THE OPERATOR TO OBTAIN AND OBSERVE THE MANUFACTURERS MATERIAL SAFETY DATA SHEETS FOR CONSUMABLE MATERIALS. INFORMATION SUCH AS, HAZARDOUS INGREDIENTS, PHYSICAL/CHEMICAL CHARACTERISTICS, FIRE, EXPLOSION, REACTIVITY, HEALTH HAZARD DATA, PRECAUTIONS AND SAFE HANDLING, USE AND CONTROL MEASURES AND ALSO TO TAKE LOCAL REGULATIONS INTO CONSIDERATION.

CAUTION: MAKE SURE THE LOCKING COMPOUND IS CONFINED TO THE THREADS.

NOTE: Torque all Bolts and Nuts as instructed in SPP-V2500-1IA Standard/Processes Manual, Chapter 70-41-00.

(1) Apply two drops of CoMat 08-026 Locking Compound to the threads of the 745-3323-501 Shoulder Bolt that attaches the 745-3321-501 Pin to the 745-3325-503 Blocker Bar Assembly. Refer to Figure 1, sheet 2.

(2) Install the 745-3321-501 Pin to the 745-3325-503 Blocker Bar Assembly with the 745-3323-501 Shoulder Bolt. Torque the Bolt to 15 lbfin (1.7 Nm).

(3) Apply two drops of CoMat 08-026 Locking Compound to the threads of the two remaining 745-3323-501 Shoulder Bolts.

(4) Install the 745-3326-501 Plunger Assembly to the 745-3325-503 Blocker Bar Assembly with the two 745-3323-501 Shoulder Bolts. Torque the Bolts to 15 lbfin (1.7 Nm).

(5) Install the MS24585-387 Spring to the 745-3326-501 Plunger Assembly, as shown.

(6) Install the 745-3325-503 Blocker Bar Assembly to the Locator Body.

(7) Remove the two AS8623E Nuts, the two SP154E Washers and the NAS6304U22 Bolts from the Door Locator. Refer to Figure 1, sheet 1.

NOTE: When you remove the two AS8623E Nuts, SP154E Washers and NAS6304U22 Bolts from the Door Locator, you must hold the Throat Washer, the Shim, the Locator Body and the Guide Bar in position.

WARNING: CoMat 01-076 SOLVENT IS FLAMMABLE AND THE VAPOUR IS HARMFUL. USE IN A WELL VENTILATED AREA. AVOID PROLONGED BREATHING OF VAPOURS OR PROLONGED OR REPEATED CONTACT WITH SKIN. HIGH CONCENTRATIONS MAY CAUSE IMPAIRED JUDGEMENT. PROTECTIVE GLOVES SHOULD BE WORN DURING USE. MAY CAUSE DERMATITIS BY REMOVING SKIN OILS.PRIOR TO USE OF THIS PRODUCT, READ THE "MATERIAL SAFETY DATA SHEET" AND FOLLOW ALL LISTED SAFETY AND HEALTH PRECAUTIONS.

(8) Use a clean, lint free cloth dampened with CoMat 01-076 Solvent to remove all oil, grease, dirt or other contaminants from the mating surfaces between the Guide Bar and the 745-3318-501 Guide Machining. Wipe the area dry with a clean, lint free cloth before the CoMat 01-076 evaporates.

WARNING: CoMat 08-024 JOINTING COMPOUND IS CLASSIFIED AS A HAZARDOUS MATERIAL WHICH MAY CAUSE INJURY OR ILLNESS IF NOT PROPERLY USED. THIS PRODUCT SHOULD BE USED ONLY IN ACCORDANCE WITH THE MANUFACTURER'S SPECIFIC SAFETY AND HEALTH RECOMMENDATIONS. PRIOR TO USE OF THIS PRODUCT, CAREFULLY READ THE APPLICABLE "MATERIAL SAFETY DATA SHEET" AND FOLLOW ALL LISTED SAFETY AND HEALTH PRECAUTIONS.

(9) Put CoMat 08-024 Jointing Compound on the mating surface of the 745-3318-501 Guide Machining as an interfay between the Guide Machining and the Guide Bar. Assemble the parts while the Jointing Compound is still wet. Refer to Figure 1, sheet 2.

(10) Install the 745-3318-501 Guide Machining to the 745-3326-501 Plunger Assembly and the Guide Bar with the two NAS6304U24 Bolts, SP154E Washers and AS8623E Nuts, as shown. Torque the Nuts to 72 lbfin (8.1Nm). Remove any excess Jointing Compound.

(11) Reidentify the 740-3000-605 Air Inlet Cowl to 740-3000-609 with a metal stamp or vibro-etch engraver. Refer to the IAE/V2500 Standard Practices/Processes Manual, Chapter 70-09-00.

(12) Reidentify the 745-3000-511 Air Inlet Cowl to 745-3000-515 with a metal stamp or vibro-etch engraver. Refer to the IAE/V2500 Standard Practices/Processes Manual, Chapter 70-09-00.

(13) For all other standards of A1 and A5 Air Inlet Cowls not referred to in step (11) or step (12) engrave this Service Bulletin number to the Mod Plates with a metal stamp or vibro-etch engraver. Refer to the IAE/V2500 Standard Practices/Processes Manual,Chapter 70-09-00.

Part B - For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

(1) Remove the two AS8623E Nuts, the two SP154E Washers and the NAS6304U24 Bolts from the Door Locator. Refer to Figure 3, sheet 1.

NOTE: When you remove the two AS8623E Nuts, SP154E Washers and NAS6304U24 Bolts from the Door Locator, you must hold the Throat Washer, the Shim, the Locator Body and the Guide Bar in position.

(2) Remove the 745-3318-501 Guide Machining and MS24585-387 Spring from the Air Inlet Cowl.

(3) Remove the 745-3325-501 Blocker Bar Assembly from the Air Inlet Cowl.

(4) Remove the Rubber Blocks and Metal Plates from the ends of the Blocker Bar Assembly.

WARNING: CoMat 01-076 SOLVENT IS FLAMMABLE AND THE VAPOUR IS HARMFUL. USE IN A WELL VENTILATED AREA. AVOID PROLONGED BREATHING OF VAPOURS OR PROLONGED OR REPEATED CONTACT WITH SKIN. HIGH CONCENTRATIONS MAY CAUSE IMPAIRED JUDGEMENT. PROTECTIVE GLOVES SHOULD BE WORN DURING USE. MAY CAUSE DERMATITIS BY REMOVING SKIN OILS.PRIOR TO USE OF THIS PRODUCT, READ THE "MATERIAL SAFETY DATA SHEET" AND FOLLOW ALL LISTED SAFETY AND HEALTH PRECAUTIONS.

(5) Use a clean, lint free cloth dampened with CoMat 01-076 Solvent to remove all old sealant, oil, grease, dirt or other contaminants from the exposed ends of the Blocker Bar Assembly and the mating surfaces between the 745-3318-501 Guide Machining and the Guide Bar. Wipe the area dry with a clean, lint free cloth before the CoMat 01-076 evaporates.

(6) Reidentify the 745-3325-501 Blocker Bar Assembly to 745-3325-503 with a vibro-etcher. Refer to the IAE V2500 Standard Practices/Processes Manual, Chapter 70-09-00.

(7) Install the 745-3325-503 Blocker Bar Assembly to the Air Inlet Cowl. Refer to Figure 3, sheet 2.

(8) Install the MS24585-387 Spring to the 745-3326-501 Plunger Assembly.

WARNING: CoMat 08-024 JOINTING COMPOUND IS CLASSIFIED AS A HAZARDOUS MATERIAL WHICH MAY CAUSE INJURY OR ILLNESS IF NOT PROPERLY USED. THIS PRODUCT SHOULD BE USED ONLY IN ACCORDANCE WITH THE MANUFACTURER'S SPECIFIC SAFETY AND HEALTH RECOMMENDATIONS. PRIOR TO USE OF THIS PRODUCT, CAREFULLY READ THE APPLICABLE "MATERIAL SAFETY DATA SHEET" AND FOLLOW ALL LISTED SAFETY AND HEALTH PRECAUTIONS.

(9) Put CoMat 08-024 Jointing Compound on the mating surface of the 745-3318-501 Guide Machining as an interfay between the Guide Machining and the Guide Bar. Assemble the parts while the Jointing Compound is still wet.

(10) Install the 745-3318-501 Guide Machining to the 745-3326-501 Plunger Assembly and the Guide Bar with the two NAS6304U24 Bolts, SP154E Washers and AS8623E Nuts, as shown. Torque the Nuts to 72 lbfin (8.1Nm). Remove any excess Jointing Compound.

C. Function Check Instructions

(1) Do Subtask 71-13-00-860-051 as instructed in A320/V2500-A1, A319/A320/A321/V2500-A5 Aircraft Maintenance Manual, Task 71-13-00-410-010.

(2) Do Subtask 71-13-00-410-059 Steps A.(1) thru A.(6). as instructed in A320/V2500-A1, A319/A320/A321/V2500-A5 Aircraft Maintenance Manual, Task 71-13-00-410-010.

(3) Make sure there is a visible gap between the Fan Cowl Doors and the Air Inlet Cowl at the 3 O'clock / 9 O'clock positions. Refer to Figure 2.

WARNING: USE A STAR-HEADED SCREWDRIVER OF SUFFICIENT LENGTH TO OPERATE THE H.O.D. MAKE SURE YOUR HANDS REMAIN CLEAR OF THE FAN COWL DOORS WHILE YOU OPERATE THE H.O.D. FAILURE TO DO SO MAY RESULT IN INJURY TO PERSONNEL.

(4) Use a Star-Headed ScrewDriver to push the 745-3321-501 Pin towards the Door Locator until the Fan Cowl Doors are released.

(5) Remove the ScrewDriver from the Fan Cowl Doors.

(6) Do Subtask 71-13-00-410-059 Steps A.(7) thru A.(12) as instructed in A320/V2500-A1, A319/A320/A321/V2500-A5 Aircraft Maintenance Manual, Task 71-13-00-410-010.

(7) Release the four latches in sequence from the front of the Cowl to the rear.

(8) Open the L/H Fan Cowl Door until the Door is disengaged from the H.O.D.

(9) Lower the Fan Cowl Door.

(10) Make sure there is a visible gap between the L/H Fan Cowl Door and the Air Inlet Cowl at the 3 O'clock position.

(11) Use a Star-Headed ScrewDriver to push the 745-3321-501 Pin towards the Door Locator until the Fan Cowl Door is released.

(12) Remove the ScrewDriver from the Fan Cowl Doors.

(13) Open the R/H Fan Cowl Door until the Door is disengaged from the H.O.D.

(14) Lower the Fan Cowl Door.

(15) Make sure there is a visible gap between the R/H Fan Cowl Door and the Air Inlet Cowl at the 9 o'clock position.

(16) Use a Star-Headed ScrewDriver to push the 745-3321-501 Pin towards the Door Locator until the Fan Cowl Doors are released.

(17) Remove the ScrewDriver from the Fan Cowl Doors.

D. Post-requisite Instructions

(1) Close the Fan Cowl Doors as instructed in the A320/V2500-A1, A319/A320/A321/V2500-A5 Aircraft Maintenance Manual, Task 71-13-00-410-010 as follows:

(a) Do Subtask 71-13-00-860-051.

(b) Do Subtask 71-13-00-410-059 Steps A.(1) thru A.(6).

WARNING: USE A STAR-HEADED SCREWDRIVER OF SUFFICIENT LENGTH TO OPERATE THE H.O.D. MAKE SURE YOUR HANDS REMAIN CLEAR OF THE FAN COWL DOORS WHILE YOU OPERATE THE H.O.D. FAILURE TO DO SO MAY RESULT IN INJURY TO PERSONNEL.

(c) Use a Star-Headed ScrewDriver to push the 745-3321-501 Pin towards the Door Locator until the Fan Cowl Doors are released.

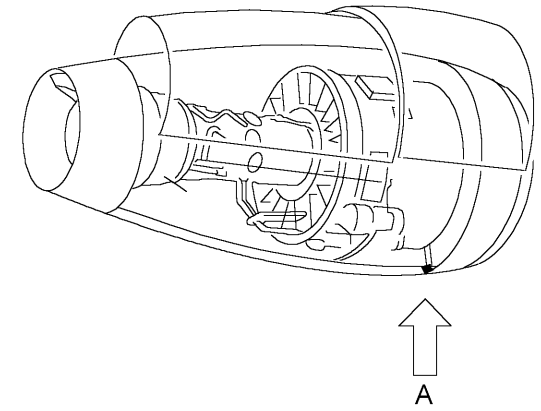
(d) Remove the Screw Driver from the Fan Cowl Doors.

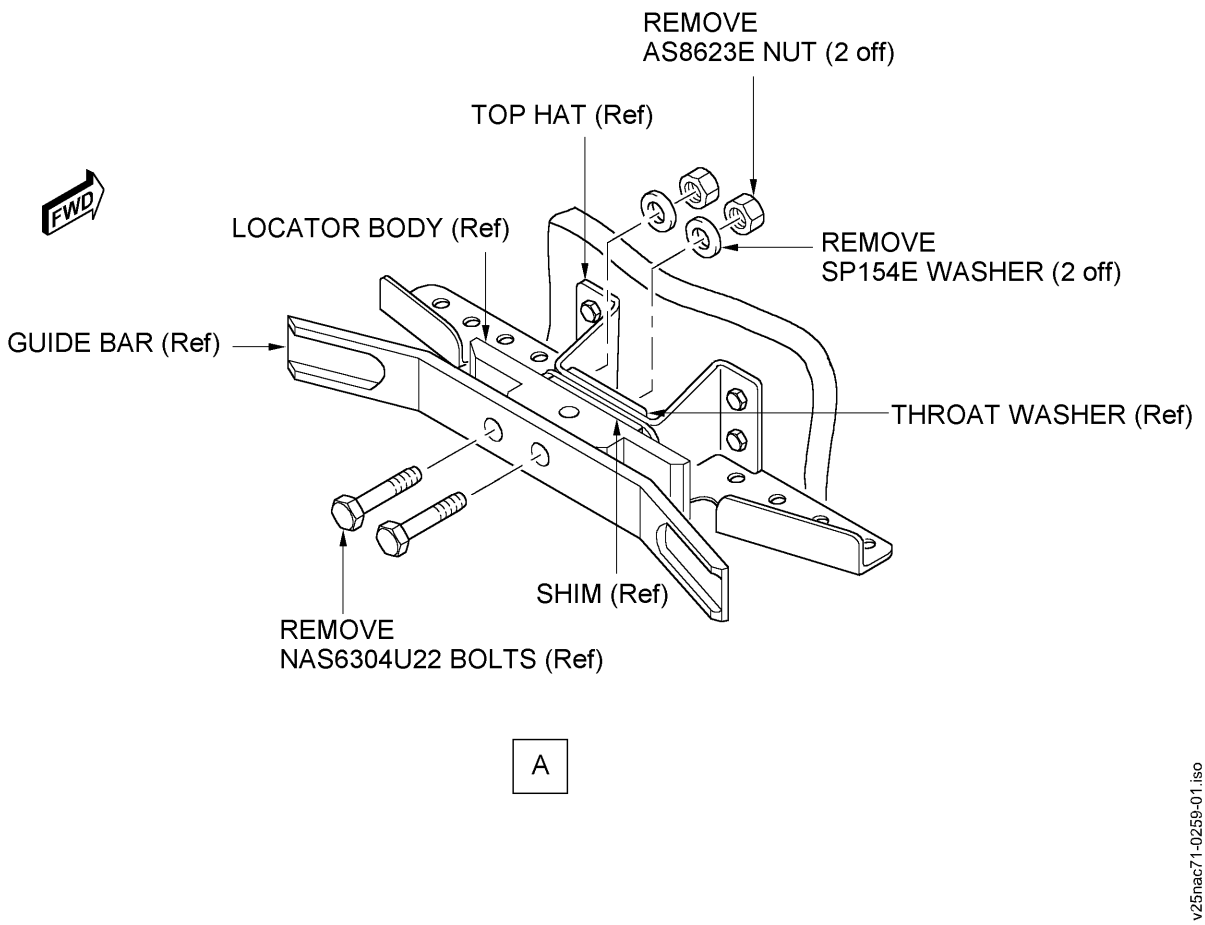
(e) Do Subtask 71-13-00-410-059 Steps A.(7) thru A.(12).

(f) Do Subtask 71-13-00-942-053.

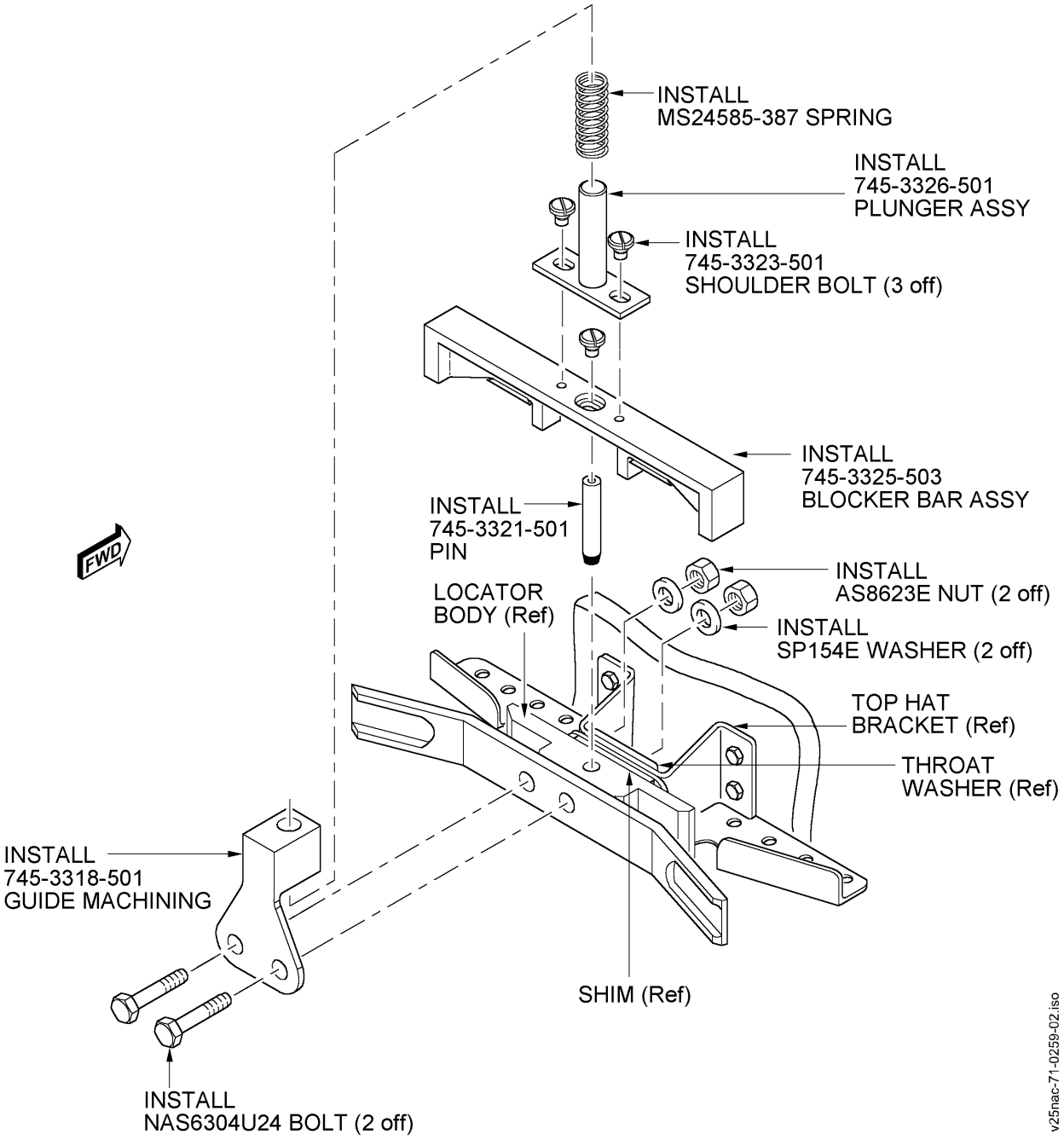
E. Recording Instructions

(1) A record of accomplishment is necessary. Write in the Aircraft Log Book that Service Bulletin V2500-NAC-71-0259 revision 3 has been done.

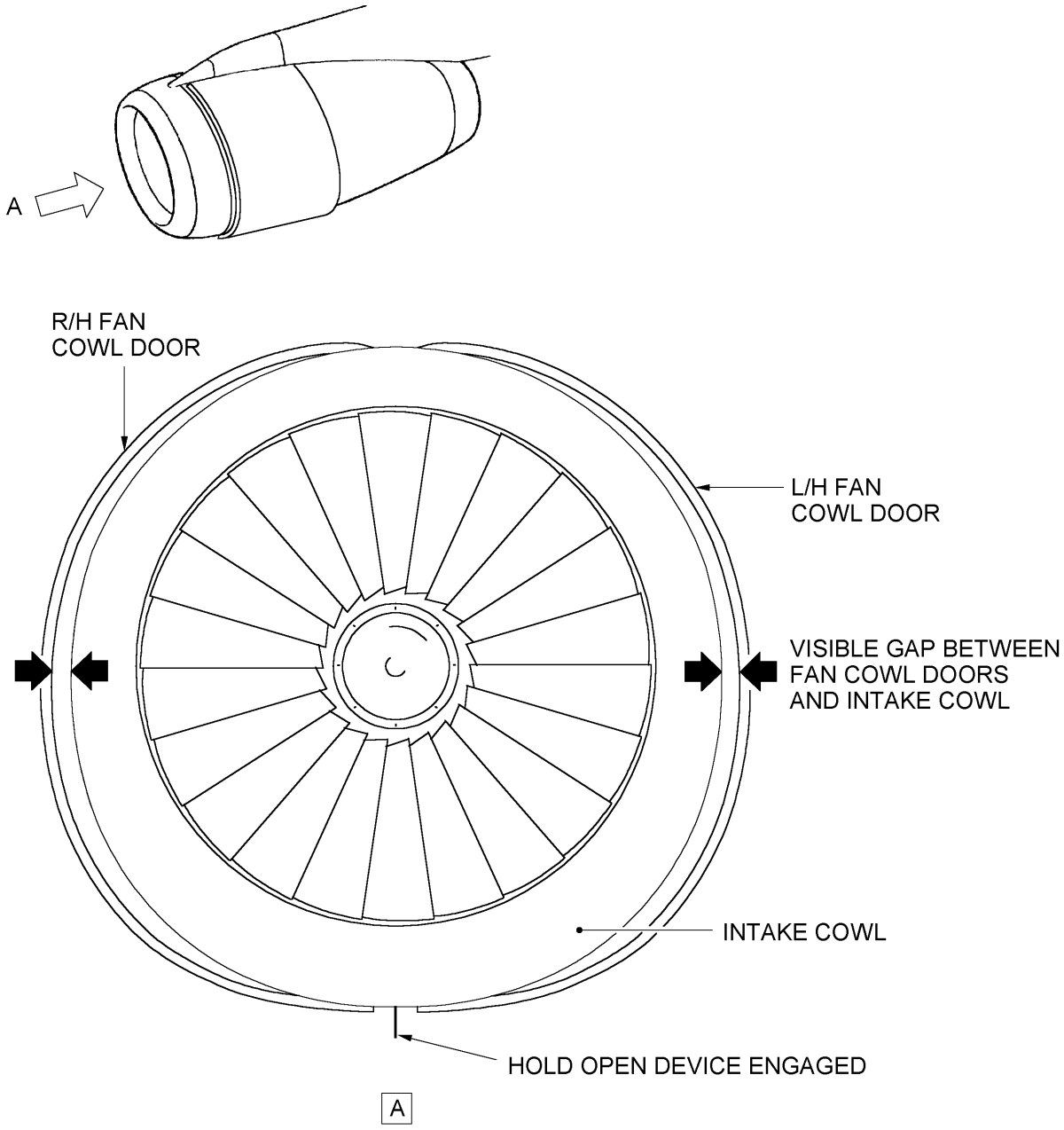




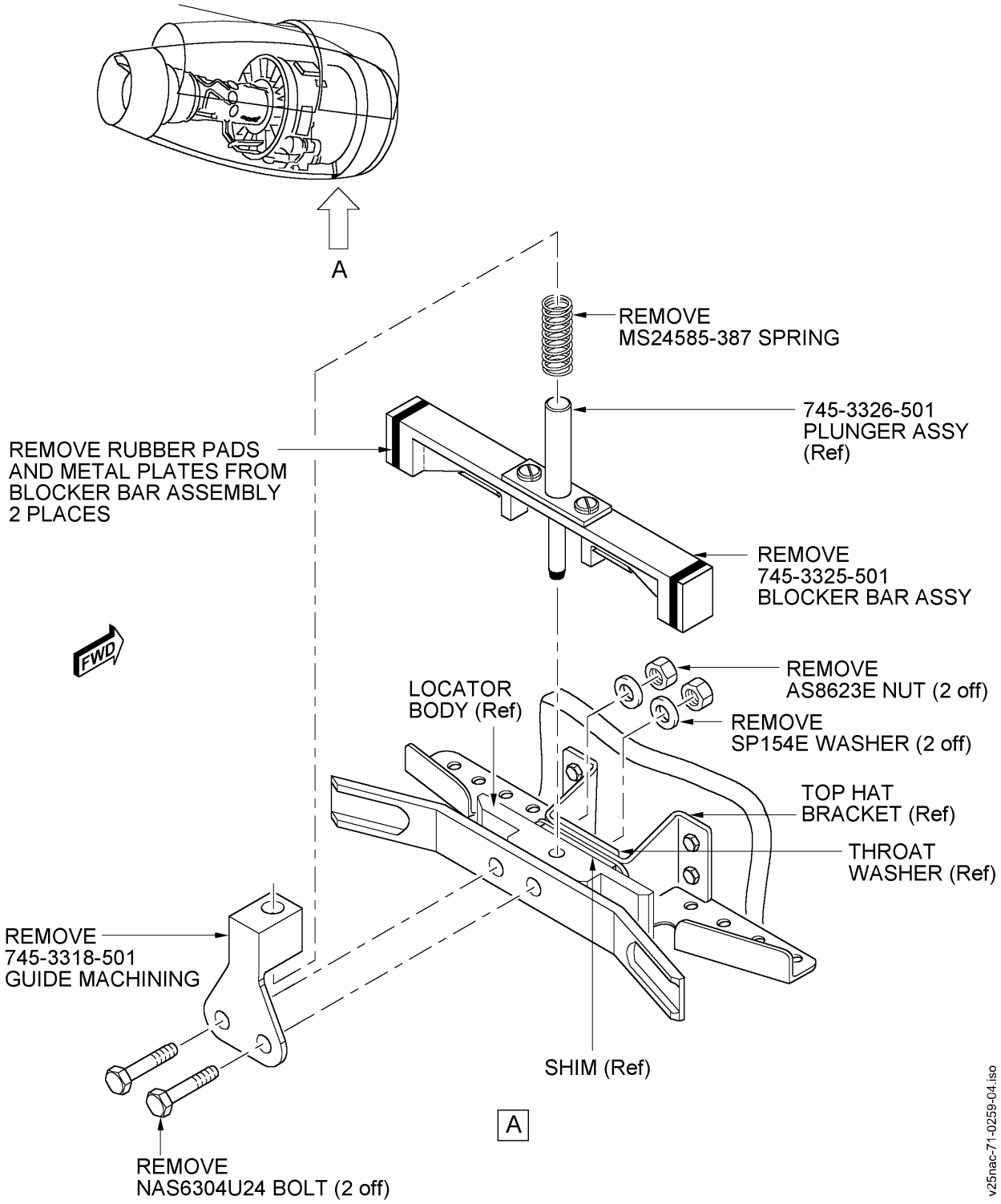
Installation of the Hold Open Device
Figure 1 Sheet 1 of 2



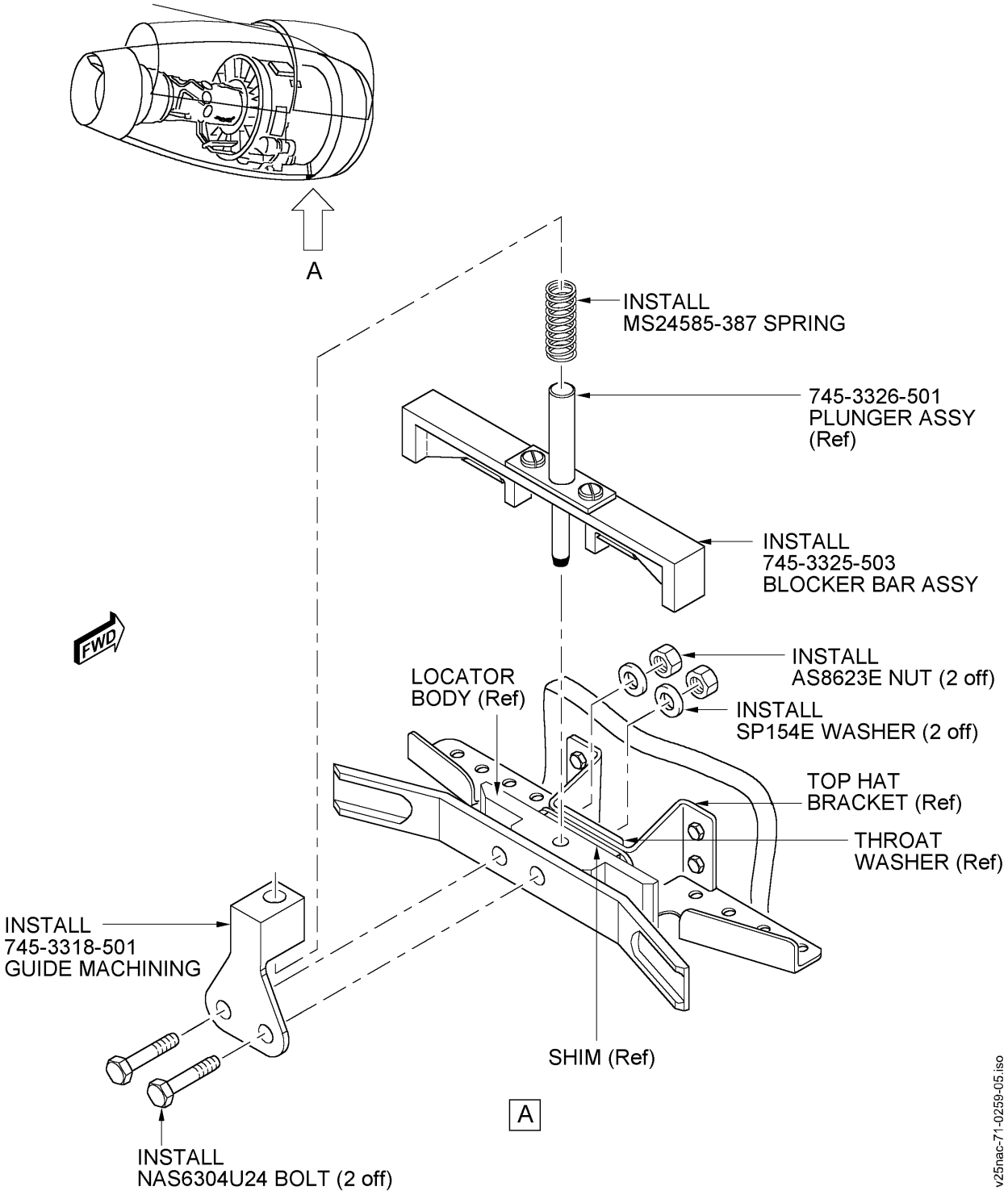
Installation of the Hold Open Device
Figure 1 Sheet 2 of 2



Functional check of the Hold Open Device
Figure 2



Removal and rework of Hold Open Device
Figure 3, sheet 1



Installation of the Hold Open Device
Figure 3, sheet 2

3. Material information

NEW PART No (ATA No)	QTY	EST'D UNIT PRICE ($)	KEYWORD	OLD PART No (IPC No)	INSTR/ DISPOS

Applicability: For each V2500-A1 or A5 Nacelle to incorporate this Bulletin.

A. Kits associated with this Bulletin

(a) For Operators who have not incorporated the original / rev 1 issue of this Service Bulletin:

V2571259-553	1	$4,313.00	Kit, Hold Open Device	(A)
Consisting of:				
745-3318-501	1		Guide Machining	
745-3321-501	1		Pin	
745-3323-501	3		Shoulder Bolt	
745-3325-503	1		Blocker Bar Assembly	
745-3326-501	1		Plunger Assembly	
MS24585-387	1		Spring	
NAS6304U24	2		Bolt	

(b) For Operators who have incorporated the original / rev 1 issue of this Service Bulletin:

None.

B. Parts affected by this Bulletin - V2500-A1

740-3000-609 (71-11-11)	1	Cowl, Air Inlet	740-3000-605 (01-005)	(B)(1D)
745-3318-501 (71-11-11)	1	.Guide Machining	--- (02-700)	(B)
745-3321-501 (71-11-11)	1	.Pin	--- (02-705)	(B)
745-3323-501 (71-11-11)	3	.Shoulder Bolt	--- (02-710)	(B)
MS24585-387 (71-11-11)	1	.Spring	--- (02-715)	(B)
745-3326-501 (71-11-11)	1	.Plunger Assembly	--- (02-720)	(B)
745-3325-503 (71-11-11)	1	.Blocker Bar Assembly	745-3325-501 (02-725)	(B)(S1) (1D)(2D)
745-3324-501 (71-11-11)	2	..Tufnol Block	--- (02-730)	(B)
MS24693-C3 (71-11-11)	4	..Screw	--- (02-735)	(B)

NAS6304U24 (71-11-11)	2	.Bolt	NAS6304U22 (02-750)	(B)

B. Parts affected by this Bulletin - V2500-A5

745-3000-515 (71-11-11)	1	Cowl, Air Inlet	745-3000-511 (01-005)	(B)(1D)
745-3318-501 (71-11-11)	1	.Guide Machining	--- (02-700)	(B)
745-3321-501 (71-11-11)	1	.Pin	--- (02-705)	(B)
745-3323-501 (71-11-11)	3	.Shoulder Bolt	--- (02-710)	(B)
MS24585-387 (71-11-11)	1	.Spring	--- (02-715)	(B)
745-3326-501 (71-11-11)	1	.Plunger Assembly	--- (02-720)	(B)
745-3325-503 (71-11-11)	1	.Blocker Bar Assembly	745-3325-501 (02-725)	(B)(S1) (1D)(2D)
745-3324-501 (71-11-11)	2	..Tufnol Block	--- (02-730)	(B)
MS24693-C3 (71-11-11)	4	..Screw	--- (02-735)	(B)
NAS6304U24 (71-11-11)	2	.Bolt	NAS6304U22 (02-750)	(B)

C. Instructions/Dispositions Code Statements

(A) Kit is currently available.
(B) New parts are currently available.
(1D) Old part number can be reworked to new part number as instructed in this Service Bulletin.
(2D) Old part is no longer available.
(S1) New parts coded (S1) and old parts coded (S1) are fully and freely interchangeable.

NOTE: Contact your BFGoodrich Regional Business Manager for pricing information for this Service Bulletin Kit.

D. Materials required to incorporate this Bulletin.

CoMat	01-076	Methylethylketone (MEK)
CoMat	02-099	Lint free cloth
CoMat	08-026	Locking Compound
CoMat	08-024	PR1422 B½ Jointing Compound

NOTE: To identify the consumable materials, refer to the Overhaul Processes and Consumable Index PCI-V2500-1IA.

V2500-NAC-71-0259 Rev 2 Incorporation Report Form	
Operator	
Aircraft Tail Number	
Air Inlet Cowl	**Serial Number**
	Part Number
	Bulletin Incorporation Date
Air Inlet Cowl	**Serial Number**
	Part Number
	Bulletin Incorporation Date

Please fax this form to:

IAE - Dave Crane - 44-1332-244067
BFGoodrich - Robert Rothwell - (619) 691-6403